



CIVIL AVIATION AUTHORITY
CZECH REPUBLIC
Airworthiness Division

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AIRWORTHINESS DIRECTIVE

Number: CAA-AD-4-094/98R1

Replaces CAA-AD-4-094-98

Date of issue: July 07, 2003

AIRBUS
A310

APPLIANCE - FUSELAGE - INSPECTION/MODIFICATION

| **Applicability:** AIRBUS A310 aircraft, all certified models and all serial numbers.

| **Effective date:** September 04, 2003

| **Compliance:** Required as indicated DGAC AD 1992-106-132(B) R6.

Remarks: The compliance of this AD must be recorded in Aircraft Logbook, where applicable the requirements of this AD must be integrated into Aircraft Technical Documentation. Address inquiries concerning this AD to: Civil Aviation Authority, Airworthiness Division, Ruzyne Airport, 160 08 Prague 6, Czech Republic, tel.: 420 2 33320922, fax: 420 2 20562270.

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Director

DGAC AD No.: 1992-106-132(B) R6

AIRBUS A310 aircraft

Structure fatigue (ATA 53, 55, 57)

APPLICABILITY:

AIRBUS A310 aircraft, all certified models and all serial numbers.

REASONS:

The goal of this Airworthiness Directive (AD) is to detect or prevent possible damage associated with a phenomenon of structural fatigue.

Revision 6 of this AD cancels part 1.11 of paragraph COMPLIANCE which is replaced by AD 2003-242(B).

COMPLIANCE:

1. Perform the inspections and/or modifications defined in the following AIRBUS Service Bulletins, in accordance with the thresholds indicated in each of these Service Bulletins or within 1000 flights following the effective date of this Airworthiness Directive Revision 3 (June 17, 1995), whichever occurs later.

1.1. S.B. A310-53-2014 Revision 5 and Change Notice 5B:
Fuselage - Center Section - Reinforce bottom joint angle fitting at FR40.

1.2. S.B. A310-53-2016 Revision 5:
Fuselage - Center Section - FR47 and FR54 area - Provide local reinforcement.

1.3. S.B. A310-53-2054 Revision 2:
Fuselage - Center Section - Inspect frame reinforcement angle runout of FR46 between STGR21 and 22 LH/RH.

1.4. S.B. A310-53-2057 Revision 1:
Fuselage - Center section - Inspection of the T-section connecting FR50A to the beam between FR50 and FR51 LH/RH.

1.5. S.B. A310-53-2059 Revision 1:
Fuselage - Inspect the lower side panel at the lap joint with the upper side panel at FR47 and STGR22, LH/RH.

1.6. S.B. A310-55-2002 Revision 4:
Stabilizers - Horizontal stabilizer structure - Reinforcement of junction parts.

1.7. S.B. A310-55-2004 Revision 3:
Stabilizers - Horizontal stabilizer structure - Inspection program after reinforcement as per Service Bulletin n° A310-55-2002 embodiment.

1.8. S.B. A310-57-2002 Revision 2:
Wings - Leading edge access panels landing - Lower skin - Inspection for cracks at bolt holes.

1.9. S.B. A310-57-2006 Revision 3:
Wings - Holes around overwing refueling aperture at RIBS 13 -14, Inspection.

1.10 S.B. A310-57-2032 Revision 3:
Wings - Upper skin forward of front spar - Inspection for cracks.

1.11 Cancelled [see AD 2003-242(B)].

1.12 S.B. A310-57-2039:
Wings - Center wing - Inspection of front spar web vertical posts.

1.13 S.B. A310-57-2046 Revision 6:

Wings - Rear spar at selected bolt locations for attachment of MLG forward pick up fitting.

1.14 S.B. A310-57-2047 Revision 3:

Wings - Center wing - Inspect rear spar internal angle and tee fitting.

1.15 S.B. A310-57-2050 and Change Notice OB:

Wings - Center wing box - Inspect treillis boom drain holes.

1.16 S.B. A310-53-2074 Revision 2:

Fuselage - Tail cone - Inspect the lower horizontal-stabilizer cutout longeron, the corner fitting, the skin stap and the skin between FR87 and FR89 and between STGR24 and STGR27, LH and RH.

1.17 S.B. A310-57-2064:

Fuselage - FR40 - Inspect upper corner fitting.

1.18 S.B. A310-57-2038 Revision 3:

Wings - Stringer flanges at rib 14 wing bottom skin - Inspect for cracks.

NOTE 1: For paragraphs 1.1, 1.10, [...], 1.17, 1.18 comply with the instructions of above-mentioned Service Bulletins at the defined thresholds or within 1000 flight following the effective date of this Airworthiness Directive Revision 4, whichever occurs later.

2. Repeat the inspections defined in the Service Bulletins mentioned in paragraph 1 above, in accordance with the intervals indicated in each of the Service Bulletins when applicable.

NOTE 2: The thresholds and intervals which are established and valid for aircraft operated for an individual average airborne flight time of 96 minutes must be adjusted in accordance with the instructions given in the Service Bulletins quoted above.

REF.: AIRBUS Service Bulletins:

A310-53-2014, -2016 -2054 -2057 -2059,-2074,

A310-55-2002, -2004

A310-57-2002, -2006, -2032, [...], -2038, -2039, -2046, -2047, -2050, -2064.

Any later approved revision of these SB's is acceptable.

This Revision 6 replaces Airworthiness Directive 92-106-132(B) R5 dated November 18, 1998.

EFFECTIVE DATES:

Original issue: MAY 09, 1992

Revision 1: MAY 09, 1992

Revision 2: JANUARY 28, 1995

Revision 3: JUNE 17, 1995

Revision 4: JUNE 15, 1996

Revision 5: NOVEMBER 28, 1998

Revision 6: JULY 05, 2003