



CIVIL AVIATION AUTHORITY
CZECH REPUBLIC
Airworthiness Division
Airport Ruzyně, 160 08 Prague 6
Tel: 420 233320922, fax: 420 220562270

AIRWORTHINESS DIRECTIVE

Number: CAA-AD-025/2003

Date of issue: February 21, 2003

**Robinson Helicopter Company
R44**

HELICOPTER – TAIL ROTOR PITCH CONTROL BEARING – INSPECTION/REPLACEMENT

Applicability: Model R44 helicopters, up to and including serial number 1208, except serial numbers 1143, 1165, 1183, 1189, 1192, 1196, 1197, 1198, 1200, 1203, and 1204, with pitch control assembly, part number (P/N) C031-1, Revision G or prior, installed, certificated in any category.

Effective date: March 26, 2003

Compliance: Required as indicated in FAA AD 2003-04-05.

Remarks: The compliance of this AD must be recorded in Aircraft Logbook, where applicable the requirements of this AD must be integrated into Aircraft Technical Documentation. Address inquiries concerning this AD to: Civil Aviation Authority, Airworthiness Division, Ruzyně Airport, 160 08 Prague 6, Czech Republic, tel: 420 233320922, fax: 420 220562270.

Ing. Pavel MATOUŠEK
director

2003-04-05 Robinson Helicopter Company: Amendment 39-13053. Docket No. 2001-SW-45-AD.

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Note 1: This AD applies to each helicopter identified in the preceding applicability provision, regardless of whether it has been otherwise modified, altered, or repaired in the area subject to the requirements of this AD. For helicopters that have been modified, altered, or repaired so that the performance of the requirements of this AD is affected, the owner/operator must request approval for an alternative method of compliance in accordance with paragraph (b) of this AD. The request should include an assessment of the effect of the modification, alteration, or repair on the unsafe condition addressed by this AD; and if the unsafe condition has not been eliminated, the request should include specific proposed actions to address it.

Compliance: Required as indicated, unless accomplished previously.

To detect corrosion of a tail rotor pitch control bearing (bearing) and to prevent bearing failure and subsequent loss of directional control of the helicopter, accomplish the following:

(a) Within 20 hours time-in-service (TIS) and thereafter at intervals not to exceed 300 hours TIS or 12 months, whichever occurs first, inspect the pitch control assembly for roughness or binding of the pitch control bearings by hand rotating the pitch control bearing housing (housing) in accordance with Robinson Helicopter Company Service Bulletin SB-43A, Revision A, dated June 10, 2002. If the housing does not rotate freely, before further flight, replace the unairworthy pitch control assembly with an airworthy unit.

(b) An alternative method of compliance or adjustment of the compliance time that provides an acceptable level of safety may be used if approved by the Manager, Los Angeles Aircraft Certification Office (LAACO), FAA. Operators shall submit their requests through an FAA Principal Maintenance Inspector, who may concur or comment and then send it to the Manager, LAACO.

Note 2: Information concerning the existence of approved alternative methods of compliance with this AD, if any, may be obtained from the LAACO.

(c) Special flight permits may be issued in accordance with 14 CFR 21.197 and 21.199 to operate the helicopter to a location where the requirements of this AD can be accomplished.

(d) The inspection of the pitch control assembly shall be done in accordance with Robinson Helicopter Company Service Bulletin SB- 43A, Revision A, dated June 10, 2002. This incorporation by reference was approved by the Director of the Federal Register in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. Copies may be obtained from Robinson Helicopter Company, 2901 Airport Drive, Torrance, California 90505, telephone (310) 539-0508, fax (310) 539- 5198. Copies may be inspected at the FAA, Office of the Regional Counsel, Southwest Region, 2601 Meacham Blvd., Room 663, Fort Worth, Texas; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC.

(e) This amendment becomes effective on March 26, 2003.