



CIVIL AVIATION AUTHORITY
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AIRWORTHINESS DIRECTIVE

Number: CAA-AD-021/2004

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**GENERAL ELECTRIC Comp.
CF6-80**

ENGINE - HIGH PRESSURE TURBINE - ROTOR DISK - INSPECTION/REPLACEMENT

Applicability: the General Electric Company (GE) CF6-80 turbofan engine models listed in Table 1 FAA AD 2004-04-07.

Effective date: April 15, 2004

Compliance: Required as indicated FAA AD 2004-04-07.

Remarks: The compliance of this AD must be recorded in Aircraft Logbook, where applicable the requirements of this AD must be integrated into Aircraft Technical Documentation. Address inquiries concerning this AD to: Civil Aviation Authority, Airworthiness Division, Ruzyně Airport, 160 08 Prague 6, Czech Republic, tel.: 420 2 33320922, fax: 420 2 20562270.

Ing. Pavel MATOUŠEK
Director

2004-04-07 General Electric Company: Amendment 39-13488. Docket No. 2004-NE-05-AD. Supersedes AD 2001-10-07, Amendment 39-12233, and AD 2003-01-05, Amendment 39-13016.

Effective Date

(a) This airworthiness directive (AD) becomes effective March 12, 2004.

Affected ADs

(b) This AD supersedes AD 2001-10-07 and AD 2003-01-05.

Applicability

(c) This AD applies to the General Electric Company (GE) CF6-80 turbofan engine models listed in the following Table 1:

Table 1.--Applicability Models, Part Numbers, Airplanes

| Models | Stage 1 high pressure turbine (HPT) rotor disk part Nos. (PNs) | Engines installed on but not limited to |
|---|--|---|
| CF6-80A, CF6-80A1, CF6-80A2, CF6-80A3 | 9234M67G22/G24/G25/G26. 9362M58G02/ G06/G07/G09. 9367M45G02/G04/G09. | Airbus A310 and Boeing 767 airplanes. |
| CF6-80C2A1, CF6-80C2A2, CF6-80C2A3, CF6-80C2A5, CF6-80C2A8, CF6- 80C2A5F, CF6-80C2B1, CF6-80C2B2, CF6-80C2B4, CF6-80C2B6, CF6- 80C2B1F, CF6-80C2B2F, CF6-80C2B4F, CF6-80C2B5F, CF6-80C2B6F, CF6-80C2B6FA, CF6-80C2B7F, CF6-80C2D1F. | 1862M23G01. 9392M23G10/G12/G21. 1531M84G02/G06/G08/G10. | Airbus A300, A310, Boeing 747, 767, and McDonnell Douglas MD11 airplanes. |
| CF6-80E1A2, CF6-80E1A4 | 1639M41P04 | Airbus A330 airplanes. |

These engines are installed on, but not limited to, the airplanes listed in Table 1 of this AD.

Unsafe Condition

(d) This AD results from the manufacturer's investigation and development of a rework procedure that chamfers the aft breakedge of the dovetail slot bottom. The actions specified in this AD are intended to detect and prevent cracks in the bottoms of the dovetail slots that could propagate to failure of the disk and cause an uncontained engine failure.

Compliance

(e) You are responsible for having the actions required by this AD performed within the compliance times specified unless the actions have already been done.

CF6-80A, -80A1, -80A2, and -80A3 Engines

Stage 1 HPT Rotor Disks, P/N 9362M58G09, With Chamfered Breakedges

(f) At the next piece-part exposure after the effective date of this AD, for stage 1 HPT rotor disks, P/N 9362M58G09, with SNs listed in Table 2 of this AD, do the following:

Table 2.--SNs of CF6-80A Series Stage 1 HPT Rotor Disk P/N 9362M58G09-- With Chamfered Breakedges

| | | | | | |
|----------|----------|----------|----------|----------|----------|
| GWN03RD7 | GWN042J3 | GWN04HRD | GWN04M9K | GWN03TKG | GWN04FW2 |
| GWN04HRE | GWN04M9L | GWN03TKH | GWN04FW3 | GWN04HRF | GWN04M9M |
| GWN03TKJ | GWN04FW4 | GWN04HRG | GWN04M9R | GWN03W3M | GWN04FW5 |
| GWN04HRH | GWN04M9T | GWN03W3N | GWN04H0M | GWN04K8N | GWN04M9W |
| GWN03W3R | GWN04HRA | GWN04M9J | | | |

(1) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE Service Bulletin (SB) No. CF6-80A S/B 72-0788, Revision 2, dated December 17, 2003, to do the inspection.

(2) For disks that have the chamfered breakedges, remark, fluorescent penetrant inspect (FPI), and eddy current inspect (ECI) the rotor disk. Use paragraph 3.A.(1)(a) through 3.A.(1)(b) of the Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0788, Revision 2, dated December 17, 2003, to remark and inspect the rotor disk and remove from service as necessary.

(3) For disks that do not have the chamfered breakedges, inspect, rework and remark the rotor disk. Use paragraph 3.A(2)(a) through 3.A(2)(b) of the Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0788, Revision 2, dated December 17, 2003, to inspect, rework, and remark the disk and remove from service as necessary.

Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 With SNs Not Listed in Table 2 of This AD

(g) For stage 1 HPT rotor disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 with SNs not listed in Table 2 of this AD, inspect, rework, and remark the disks using paragraphs 3.A.(2) through 3.B.(2) of Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0788, Revision 2, dated December 17, 2003, at the following:

(1) For stage 1 HPT rotor disks not installed in engines with both new and old hardware, inspect, rework, remark, and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before the effective date of this AD using any version of GE SB No. CF6-80A S/B 72-0779, inspect, rework, remark, and remove from service as necessary at the next Engine Shop Visit (ESV) using the compliance times in the following Table 3:

Table 3.--Compliance Times For Inspection and Rework of CF6-80A Series Stage 1 HPT Rotor Disks, P/Ns 9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 With SNs Not Listed in Table 2 of This AD--Previously Inspected

| Stage 1 HPT rotor disk cycles-since-last-inspection (CSLI) on the effective date of this AD | Compliance time for inspection and rework |
|---|--|
| (i) More than 1,500 CSLI | At the next ESV after the effective date of this AD, but not to exceed 4,500 CSLI. |
| (ii) 1,500 CSLI or fewer | At the next ESV after the effective date of this AD, but not to exceed 3,500 CSLI. |

(3) For stage 1 HPT rotor disks that have not been inspected before the effective date of this AD using any version of GE SB No. CF6-80A S/B 72-0779, inspect, rework, remark, and remove from service as necessary at the next ESV using the compliance times in the following Table 4:

Table 4. Compliance Times for Inspection and Rework of CF6-80A Series Stage 1 HPT Rotor Disks, P/Ns

9234M67G22, G24, G25, G26, 9367M45G04, G09, 9362M58G02, G06, G07, and 9362M58G09 With SNs Not Listed in Table 2 of This AD--Not Previously Inspected

| Stage 1 HPT rotor disk cycles-since-new (CSN) on the effective date of this AD | Compliance time for inspection and rework |
|---|--|
| (i) 10,000 or more CSN | At the next ESV or within 1,000 cycles-in-service (CIS) after the effective date of this AD, whichever occurs first. |
| (ii) 5,000 or more CSN but fewer than 10,000 CSN | At the next ESV or within 2,400 CIS after the effective date of this AD, whichever occurs first, but before accumulating 11,000 CSN. |
| (iii) Fewer than 5,000 CSN | At the next ESV or within 3,500 CIS after the effective date of this AD, whichever occurs first, but before accumulating 7,400 CSN. |

Stage 1 HPT Rotor Disks, P/N 9367M45G02

(h) For stage 1 HPT rotor disks, P/N 9367M45G02, inspect the rotor disk dovetail slot bottoms and remove the disk from service as necessary using paragraphs 3.A. through 3.C.(10)(i) of Accomplishment Instructions of GE SB No. CF6-80A S/B 72-0779, Revision 1, dated January 22, 2004, at the following times:

(1) For stage 1 HPT rotor disks not installed in engines with both new and old hardware, inspect and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before the effective date of this AD using any version of GE SB No. CF6-80A S/B 72-0779, and had more than zero CSN at the time of that inspection, inspect and remove from service as necessary at each piece-part exposure.

(3) For stage 1 HPT rotor disks that have not been inspected, or were only inspected with zero CSN before the effective date of this AD using any version of GE SB No. CF6-80A S/B 72-0779, inspect and remove from service as necessary at the next ESV using the compliance times in the following Table 5:

Table 5. Compliance Times for Inspection of CF6-80A Series Stage 1 HPT Rotor Disks, P/N 9367M45G02--Not Previously Inspected

| Stage 1 HPT rotor disk CSN on the effective date of this AD | Compliance time for initial inspection |
|--|--|
| (i) 10,000 or more CSN | At the next ESV or within 1,000 CIS after the effective date of this AD, whichever occurs first. |
| (ii) 5,000 or more CSN but fewer than 10,000 CSN | At the next ESV or within 2,400 CIS after the effective date of this AD, whichever occurs first, but before accumulating 11,000 CSN. |
| (iii) Fewer than 5,000 CSN | At the next ESV or within 3,500 CIS after the effective date of this AD, whichever occurs first, but before accumulating 7,400 CSN. |

(4) Thereafter, inspect at each piece-part exposure, and remove the rotor disk from service if necessary.

CF6-80C2 Series Engines

Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges

(i) At the next piece-part exposure after the effective date of this AD, for stage 1 HPT rotor disks, P/N 1531M84G10, with SNs listed in Table 6 of this AD, do the following:

Table 6.--SNs of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1531M84G10, With Chamfered Breakedges

| | | | | | |
|----------|----------|----------|----------|----------|----------|
| GWN03111 | GWN0369J | GWN03K3F | GWN03RPD | GWN049JM | GWN03114 |
| GWN036JG | GWN03K3G | GWN03RPE | GWN049M7 | GWN03501 | GWN036JH |
| GWN03K3H | GWN03RPF | GWN049M8 | GWN03699 | GWN036JJ | GWN03K3K |
| GWN03RPG | GWN049M9 | GWN03752 | GWN036JK | GWN03K3L | GWN0402A |
| GWN04AEP | GWN03753 | GWN036JL | GWN03K3M | GWN0402E | GWN04AER |
| GWN03754 | GWN036JM | GWN03K3N | GWN0402F | GWN04AET | GWN03755 |
| GWN036JN | GWN03K3R | GWN0402G | GWN04ALR | GWN03756 | GWN0375A |
| GWN03K3T | GWN0402H | GWN04ALT | GWN03757 | GWN0375C | GWN03K3W |
| GWN0402J | GWN04ALW | GWN03759 | GWN0375D | GWN03K40 | GWN0402K |
| GWN04AM0 | GWN03981 | GWN0375E | GWN03K6J | GWN0402L | GWN04AM1 |
| GWN03982 | GWN037H2 | GWN03K7R | GWN0402M | GWN04AM2 | GWN03983 |
| GWN0398A | GWN03K7T | GWN0402N | GWN04AM3 | GWN03984 | GWN0398C |
| GWN03KR1 | GWN0402P | GWN04AM4 | GWN03985 | GWN039PF | GWN03KR2 |
| GWN040R5 | GWN04CGJ | GWN03986 | GWN039PG | GWN03KR3 | GWN0418A |
| GWN04CGL | GWN03987 | GWN039PH | GWN03KR4 | GWN0418C | GWN04CGN |
| GWN03988 | GWN039PJ | GWN03KR5 | GWN0418D | GWN04CGT | GWN03989 |
| GWN039PK | GWN03KR6 | GWN0418E | GWN04CGW | GWN04026 | GWN039PL |
| GWN03KR7 | GWN0418F | GWN04CH3 | GWN04027 | GWN039PM | GWN03KR8 |
| GWN0418G | GWN04CH5 | GWN04028 | GWN039PN | GWN03KRA | GWN0418H |
| GWN04CH8 | GWN04029 | GWN03A4J | GWN03KRC | GWN0418J | GWN04CH9 |
| GWN04189 | GWN03A4K | GWN03KRD | GWN0418K | GWN04CHA | GWN04190 |
| GWN03A4L | GWN03L2D | GWN0418L | GWN04CHC | GWN04191 | GWN03A4M |
| GWN03L2E | GWN0418M | GWN04D52 | GWN04366 | GWN03A4N | GWN03L2F |
| GWN0418N | GWN04D54 | GWN04722 | GWN03A4P | GWN03LNF | GWN0418P |
| GWN04D55 | GWN04726 | GWN03A4R | GWN03LNJ | GWN0418R | GWN04D56 |
| GWN04729 | GWN03A4T | GWN03LNK | GWN0418T | GWN04D57 | GWN031N2 |
| GWN03A4W | GWN03M88 | GWN0418W | GWN04D58 | GWN031N3 | GWN03C12 |
| GWN03M89 | GWN044DP | GWN04D59 | GWN031N4 | GWN03C13 | GWN03M8C |
| GWN0454E | GWN04DPW | GWN031N5 | GWN03C14 | GWN03M8D | GWN0454F |
| GWN04DR4 | GWN031N6 | GWN03CA0 | GWN03M8E | GWN0454G | GWN04DR9 |
| GWN031N7 | GWN03DC9 | GWN03M8F | GWN0454H | GWN04DRE | GWN031N8 |
| GWN03DCA | GWN03M8J | GWN0454J | GWN04DRJ | GWN031N9 | GWN03DCC |
| GWN03M8K | GWN0454K | GWN04E9K | GWN031NA | GWN03DCD | GWN03NHN |
| GWN0454L | GWN04E9L | GWN031NC | GWN03DCE | GWN03NHP | GWN0454M |
| GWN04E9M | GWN032G1 | GWN03DCF | GWN03NHR | GWN0454N | GWN04E9N |
| GWN032G2 | GWN03DCG | GWN03NHT | GWN045T0 | GWN04EM5 | GWN032G3 |
| GWN03DCH | GWN03R73 | GWN045T1 | GWN04EMA | GWN032G4 | GWN03DCJ |
| GWN03R74 | GWN045T2 | GWN04EMK | GWN032G5 | GWN03DCK | GWN03R75 |
| GWN045T3 | GWN04EML | GWN032G6 | GWN03DCL | GWN03R76 | GWN045T4 |
| GWN04EMM | GWN032G7 | GWN03DCM | GWN03R77 | GWN045T5 | GWN04F8N |

| | | | | | |
|----------|----------|----------|----------|----------|----------|
| GWN032G8 | GWN03DCN | GWN03R78 | GWN045T6 | GWN04F8P | GWN032G9 |
| GWN03DCP | GWN03R79 | GWN045T7 | GWN04FTJ | GWN032GE | GWN03DCR |
| GWN03R7A | GWN045T8 | GWN04FTL | GWN0335P | GWN03DME | GWN03R7C |
| GWN045T9 | GWN04FTM | GWN0335R | GWN03DMF | GWN03R7D | GWN045TA |
| GWN04FTN | GWN033C5 | GWN03ER7 | GWN03R7E | GWN045TC | GWN034KR |
| GWN03ER8 | GWN03R7F | GWN045TD | GWN034KT | GWN03ER9 | GWN03R7G |
| GWN045TE | GWN0350M | GWN03ERA | GWN03R7H | GWN045TF | GWN0350N |
| GWN03FTN | GWN03R9G | GWN045TG | GWN0350P | GWN03FTP | GWN03R9H |
| GWN045TH | GWN0350R | GWN03FTR | GWN03R9J | GWN046F6 | GWN0350T |
| GWN03FTT | GWN03R9K | GWN046F7 | GWN0350W | GWN03FTW | GWN03R9L |
| GWN046F8 | GWN035M5 | GWN03FW0 | GWN03R9M | GWN047LG | GWN035M6 |
| GWN03H56 | GWN03R9N | GWN047LH | GWN035M7 | GWN03H57 | GWN03R9P |
| GWN047LJ | GWN035M8 | GWN03H58 | GWN03R9R | GWN047LK | GWN035M9 |
| GWN03HTL | GWN03R9T | GWN047LL | GWN035MA | GWN03HTM | GWN03R9W |
| GWN048CD | GWN035MC | GWN03HTN | GWN03RA0 | GWN048CF | GWN035MD |
| GWN03HTP | GWN03RA1 | GWN048CG | GWN035TH | GWN03HTR | GWN03RA2 |
| GWN048CH | GWN035TJ | GWN03HTT | GWN03RA3 | GWN048CJ | GWN035TK |
| GWN03J8T | GWN03RA4 | GWN048CK | GWN035TL | GWN03J8W | GWN03RA5 |
| GWN048CM | GWN035TM | GWN03J90 | GWN03RA6 | GWN048CN | GWN0369A |
| GWN03J91 | GWN03RA7 | GWN048CP | GWN0369C | GWN03J92 | GWN03RA8 |
| GWN048CR | GWN0369D | GWN03JNN | GWN03RP7 | GWN049GH | GWN0369E |
| GWN03JNP | GWN03RP9 | GWN049GJ | GWN0369G | GWN03K3C | GWN03RPA |
| GWN049GK | GWN0369H | GWN03K3D | GWN03RPC | GWN049JL | |

(1) Visually inspect the rotor disks for the presence of a chamfer on the aft breakedges of the dovetail slot bottoms. Use paragraph 3.A. of GE SB No. CF6-80C2 S/B 72-1089, Revision 2, dated December 18, 2003, to do the inspection.

(2) For disks that have the chamfered breakedges, remark, FPI, and ECI the rotor disk. Use paragraph 3.A.(1)(a) through 3.A.(1)(b) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1089, Revision 2, dated December 18, 2003, to remark and inspect the rotor disk, and remove from service as necessary.

(3) For disks that do not have the chamfered breakedges, inspect, rework and remark the rotor disk. Use paragraph 3.A.(2)(a) through 3.A.(2)(b) of the Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1089, Revision 2, dated December 18, 2003, to inspect, rework and remark the disk and remove from service as necessary.

Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 With SNs Not Listed in Table 6 of This AD

(j) For stage 1 HPT rotor disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 with SNs not listed in Table 6 of this AD, inspect, rework, and remark the disks using paragraphs 3.A.(2) through 3.B.(2) of Accomplishment Instructions of GE SB No. CF6-80C2 S/B 72-1089, Revision 2, dated December 18, 2003, at the following:

(1) For stage 1 HPT rotor disks not installed in engines with both new and old hardware, inspect, rework, remark, and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before the effective date of this AD using GE SB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000, or any version of GE ASB No. CF6-80C2 S/B 72-

A1026, inspect, rework, remark, and remove from service as necessary at the next ESV using the compliance times in the following Table 7:

Table 7.--Compliance Times for Inspection and Rework of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 With SNs Not Listed in Table 6 of This AD-- Previously Inspected

| Stage 1 HPT rotor disk cycles-since-inspection (CSI) on the effective date of this AD | Compliance time for inspection and rework |
|--|---|
| (i) More than 1,500 CSLI | At the next ESV or within 4,500 CSLI after the effective date of this AD, whichever occurs first. |
| (ii) 1,500 CSLI or fewer | At the next ESV or within 3,500 CSLI after the effective date of this AD, whichever occurs first. |

(3) For stage 1 HPT rotor disks that have not been inspected before the effective date of this AD using GE SB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000, or any version of GE ASB No. CF6-80C2 S/B 72-A1026, inspect, rework, remark, and remove from service as necessary at the next ESV using the compliance times in the following Table 8:

Table 8.--Compliance Times for Inspection and Rework of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/Ns 9392M23G10, G12, G21, 1531M84G02, G06, G08, and 1531M84G10 With SNs Not Listed in Table 6 of This AD--Not Previously Inspected

| Stage 1 HPT rotor disk cycles-since-new (CSN) on the effective date of this AD | Compliance time for inspection and rework |
|---|---|
| (i) 10,000 or more CSN | At the next ESV or within 1,000 CIS after the effective date of this AD, whichever occurs first. |
| (ii) 5,000 or more CSN but fewer than 10,000 CSN | At the next ESV or within 2400 CIS after the effective date of this AD, whichever occurs first, but before accumulating 11,000 CSN |
| (iii) Fewer than 5,000 CSN. | At the next ESV or within 3,500 CIS after the effective date of this AD, whichever occurs first, but before accumulating 7,400 CSN. |

Stage 1 HPT Rotor Disks, P/N 1862M23G01

(k) For stage 1 HPT rotor disk, P/N 1862M23G01, inspect the rotor disk dovetail slot bottoms and remove the disk from service as necessary using paragraphs 3.A. through 3.C.(10)(i) of Accomplishment Instructions of GE ASB No. CF6-80C2 S/B 72-A1026, Revision 2, dated January 22, 2004, at the following times:

(1) For stage 1 HPT rotor disks not installed in engines with both new and old hardware, inspect and remove from service as necessary before further flight.

(2) For stage 1 HPT rotor disks that have been inspected before the effective date of this AD using any version of GE ASB No. CF6- 80C2 S/B 72-A1026, and had more than zero CSN at the time of that inspection, inspect and remove from service as necessary at each piece-part exposure.

(3) For stage 1 HPT rotor disks that have not been inspected, or were only inspected with zero CSN before the effective date of this AD using any version of GE ASB No. CF6-80C2 S/B 72-A1026, inspect and remove from service as necessary at the next ESV using the compliance times in the following Table 9:

Table 9.--Compliance Times for Inspection of CF6-80C2 Series Stage 1 HPT Rotor Disks, P/N 1862M23G01-- Not Previously Inspected

| Stage 1 HPT rotor disk CSN on the effective date of this AD | Compliance time for initial inspection |
|---|--|
| (i) 10,000 or more CSN | At the next ESV or within 1,000 CIS after the effective date of this AD, whichever occurs first. |
| (ii) 5,000 or more CSN but fewer than 10,000 CSN | At the next ESV or within 2,400 CIS after the effective date of this AD, whichever occurs first, but before accumulating 11,000 CSN. |
| (iii) Fewer than 5,000 CSN | At the next ESV or within 3,500 CIS after the effective date of this AD, whichever occurs first, but before accumulating 7,400 CSN. |

(4) Thereafter, inspect at each piece-part exposure, and remove the rotor disk from service if necessary.

CF6-80E1A2, A4 Engines

Stage 1 HPT Rotor Disks, P/N 1639M41P04

(1) For stage 1 HPT rotor disks, P/N 1639M41P04, remove the rotor disks from service using paragraphs 3.A.(1) through 3.A.(2) of Accomplishment Instructions of GE SB No. CF6-80E1 S/B 72-0251, dated January 22, 2004, at the following times:

(1) For stage 1 HPT rotor disks currently in service, remove the disk using the compliance times in the following Table 10:

Table 10.--Compliance Times for Removal of CF6-80E1 Stage 1 HPT Rotor Disks, P/N 1639M41P04

| Stage 1 HPT rotor disk CSN on the effective date of this AD | Compliance time for removal of disk |
|--|--|
| (i) More than 10,000 CSN | At the next ESV or within 600 CIS after the effective date of this AD, whichever occurs first. |
| (ii) More than 5,000 CSN but fewer than or equal to 10,000 CSN | At the next ESV or within 2,500 CIS after the effective date of this AD, whichever occurs first, but before accumulating 10,600 CSN. |
| (iii) Fewer than or equal to 5,000 CSN | At the next ESV or within 3,500 CIS after the effective date of this AD, whichever occurs first, but before accumulating 7,500 CSN. |

(2) After the effective date of this AD, do not install any stage 1 HPT rotor disk, P/N 1639M41P04, into any engine.

Definitions

(m) For the purpose of this AD, the following definitions apply:

(1) An engine shop visit (ESV) is defined as the removal of an engine from an aircraft for maintenance in which a major engine flange is disassembled after the effective date of this AD. The following actions, either separately or in combination with each other, are not considered ESVs for the purpose of this AD.

(i) The removal of the upper compressor stator case solely for airfoil maintenance.

(ii) The module level inspection of the high-pressure compressor rotor 3-9 spool.

- (iii) The replacement of stage 5 high-pressure compressor variable stator vane bushings or lever arms.
 (2) Piece-part exposure is defined as when:

(i) The stage 1 HPT rotor disk is considered completely disassembled according to the manufacturer's engine manual or other FAA-approved engine manual; and

(ii) The disk has accumulated more than 100 cycles-in-service since the last piece-part inspection, provided that the part was not damaged or the disassembly is not related to the cause for its removal from the engine.

Reporting Requirements

(n) Within five calendar days of the inspection, report the results of inspections that equal or exceed the reject criteria to: Engine Certification Office, FAA, Engine and Propeller Directorate, 12 New England Executive park, Burlington, MA 01803-5299; telephone (781) 238-7128; fax (781) 238-7199. Reporting requirements have been approved by the Office of Management and Budget and assigned OMB control number 2120-0056. Be sure to include the following information:

- (1) Engine model in which the stage 1 HPT rotor disk was installed.
- (2) Part Number.
- (3) Serial Number.
- (4) Part CSN.
- (5) Part CSLI.
- (6) Date and location where inspection was done.

(o) We recommend that you record the inspection information and results on GE Form 1653-1, entitled CF6-80A/80C Stage 1 HPT Disk Dovetail Slot Bottom Inspection. This form is available in any version of GE SB CF6-80A S/B 72-0779, or GE ASB CF6-80C2 S/B 72- A1026. We also recommend that a copy of the data be sent to GE Airline Support Engineering, General Electric Aircraft Engines, Customer Support Center, 1 Neumann Way, Mail Drop RM285, Cincinnati, OH, 45215.

Alternative Methods of Compliance

(p) The manager, Engine Certification Office, has the authority to approve alternative methods of compliance for this AD if requested using the procedures found in 14 CFR 39.19.

Material Incorporated by Reference

(q) You must use the service information specified in Table 11 to perform the actions required by this AD. The Director of the Federal Register approved the incorporation by reference of the documents listed in Table 11 of this AD in accordance with 5 U.S.C. 552(a) and 1 CFR part 51. You can get a copy from General Electric Company via Lockheed Martin Technology Services, 10525 Chester Road, Suite C, Cincinnati, Ohio 45215, telephone (513) 672-8400, fax (513) 672-8422. You may review copies at the FAA, New England Region, Office of the Regional Counsel, 12 New England Executive Park, Burlington, MA; or at the Office of the Federal Register, 800 North Capitol Street, NW., suite 700, Washington, DC. Table 11 follows:

Table 11.--Incorporation by Reference

| Service bulletin no. | Page | Revision | Date |
|---|------|----------|--------------------|
| GE SB No. CF6-80E1 S/B 72-0251 Total Pages: 4 | All | Original | January 22, 2004. |
| GE SB No. CF6-80A S/B 72-0779 Total Pages: 34 | All | 1 | January 22, 2004. |
| GE SB No. CF6-80A S/B 72-0788 Total Pages: 10 | All | 2 | December 17, 2003. |

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|--|-----|---|--------------------|
| GE ASB No. CF6-80C2 S/B 72-A1026 Total Pages: 38 | All | 2 | January 22, 2004. |
| GE SB No. CF6-80C2 S/B 72-1089 Total Pages: 11 | All | 2 | December 18, 2003. |

Related Information

(r) GE SB No. CF6-80C2 S/B 72-A1024, Revision 1, dated November 3, 2000 also pertains to the subject of this AD.

▼ Footer Information

Issued in Burlington, Massachusetts, on February 13, 2004.
Peter A. White,
Acting Manager, Engine and Propeller Directorate,
Aircraft Certification Service.
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